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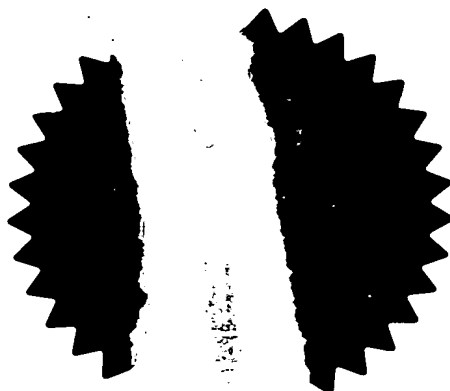
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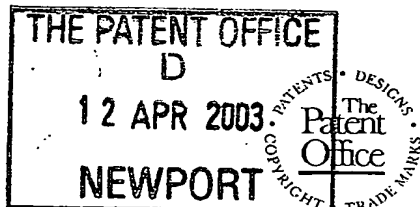
Stephen Hordley

Dated

23 January 2004

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OLIFF & BERRIDGE, PLC
P.O. BOX 19928
ALEXANDRIA, VA 22320
(703) 836-6400
APPLICANT: Timothy E. MACFADYEN
APPLICATION NO.: New U.S. Application
FILED: March 18, 2004
FOR: AIR COOLED BEARING
ATTORNEY DOCKET NO.: 118834



Request for grant of a patent

(See the notes on the back of this form. You can also get an explanatory leaflet from the Patent Office to help you fill in this form)

12 APR 2003

The Patent Office

Cardiff Road
Newport
South Wales
NP10 8QQ

1. Your reference PAT/GPW/3122

2. Patent application number
(The Patent Office will fill in this part)

0308488.6

3. Full name, address and postcode of the or of each applicant (underline all surnames)

ROLLS-ROYCE plc
65 BUCKINGHAM GATE
LONDON
SW1E 6AT
GREAT BRITAIN

Patents ADP number (if you know it)

If the applicant is a corporate body, give the country/state of its incorporation

GREAT BRITAIN

0003970002

4. Title of the invention AIR COOLED BEARING

5. Name of your agent (if you have one)

MR V J BIRD
INTELLECTUAL PROPERTY DEPT WH 58
ROLLS-ROYCE plc
PO BOX 3
FILTON
BRISTOL
BS34 7QE

"Address for service" in the United Kingdom to which all correspondence should be sent (including the postcode)

Patents ADP number (if you know it)

7546443002

6. If you are declaring priority from one or more earlier patent applications, give the country and the date of filing of the or of each of these earlier applications and (if you know it) the or each application number

Country

Priority application number
(if you know it)

Date of filing
(day / month / year)

7. If this application is divided or otherwise derived from an earlier UK application, give the number and the filing date of the earlier application

Number of earlier application

Date of filing
(day / month / year)

8. Is a statement of inventorship and of right to grant of a patent required in support of this request? (Answer 'Yes' if:

YES

- a) any applicant named in part 3 is not an inventor, or
 - b) there is an inventor who is not named as an applicant, or
 - c) any named applicant is a corporate body.
- See note (d))

Patents Form 1/77

9. Enter the number of sheets for any of the following items you are filing with this form. Do not count copies of the same document

Continuation sheets of this form 0

Description 5

Claim(s) 2

Abstract 0

Drawing(s) 3

10. If you are also filing any of the following, state how many against each item.

Priority documents 0

Translations of priority documents 0

Statement of inventorship and right to grant of a patent (Patents Form 7/77) YES

Request for preliminary examination and search (Patents Form 9/77) YES

Request for substantive examination (Patents Form 10/77) NO

Any other documents (please specify) NO

11. I/We request the grant of a patent on the basis of this application.

Signature

V J BIRD

Date

11 APRIL 2003

12. Name and daytime telephone number of person to contact in the United Kingdom PAUL WALKER 0117 979 7416
paul.walker@rolls-royce.com

Warning

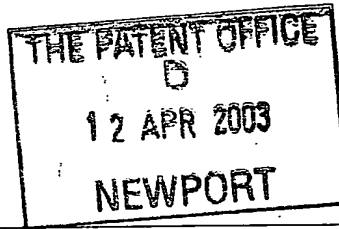
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Statement of inventorship and of right to grant of a patent



The Patent Office

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NP10 8QQ

1. Your reference PAT/GPW/3122

2. Patent application number
(if you know it)

3. Full name of the or of each applicant ROLLS-ROYCE plc

4. Title of the invention
AIR COOLED BEARING

5. State how the applicant(s) derived the right from the inventor(s) to be granted a patent BY VIRTUE OF THE INVENTORS' TERMS AND CONDITIONS OF EMPLOYMENT

6. How many, if any, additional Patents Forms 7/77 are attached to this form? NO
(see note (c))

7. I/We believe that the person(s) named over the page (and on any extra copies of this form) is/are the inventor(s) of the invention which the above patent application relates to.

Signature

Date

V J BIRD

11 APRIL 2003

8. Name and daytime telephone number of person to contact in the United Kingdom PAUL WALKER 0117 979 7416
paul.walker@rolls-royce.com

Notes

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Enter the full names, addresses and postcodes of the inventors in the boxes and underline the surnames

TIMOTHY ERIC MACFADYEN
Highbank
Selsley Road
North Woodchester
Stroud
Gloucestershire
GL5 5PH

8609174001

Patents ADP number (if you know it):

Patents ADP number (if you know it):

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Reminder

Have you signed the form?

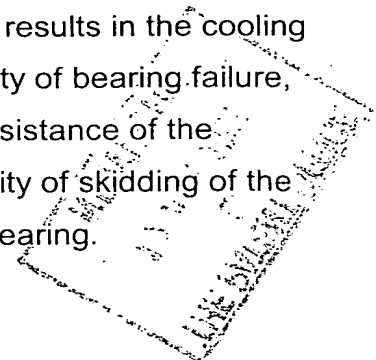
AIR COOLED BEARING

This invention relates to an air cooled bearing, and in particular an air cooled bearing in a gas turbine engine where the bearing is cooled by a continuous stream of cooling air.

The main engine bearings in a gas turbine engine are usually fed with a continuous flow of oil which not only lubricates the but also cools the bearings in use. Total loss oil systems are used in engines such as the Rolls Royce Viper turbojet engine where the bearings are fed approximately 250mls of oil per hour. The relatively low oil flow rate lubricates but does not cool the bearings. Total loss lubrication systems are particularly suitable for engines which run for short periods of time, such as vertical lift engines or expendable engines that have a relatively short design life. Total loss oil systems are much simpler than recirculatory type oil systems since it is possible to dispense with oil coolers, scavenge pumps and/or filters. In this type of lubrication system once the oil has lubricated the bearing the oil is either injected into the main gas stream of the engine or retained within an oil reservoir which drains overboard when the engine is shut down.

Bearings in engines having a total loss oil system generally run hotter than those in engines where the bearings are cooled by a continuous flow of oil used for lubrication. The bearings are usually cooled using relatively large amounts of cooling air bled from the engine compressor in use.

In one arrangement cooling air enters the bearing flowing in the axial direction of the bearing such that the cooling air passes through the gap between the cage and the rolling elements of the bearing. This has the effect that the cage and rolling elements impart work on the cooling air as it passes through the gap which results in the cooling air and the bearing being heated, thereby increasing the probability of bearing failure, and also results in a torque being applied to the bearing by the resistance of the cooling air on the bearing cage. The latter increases the probability of skidding of the rolling elements with respect to the inner and outer races of the bearing.



There is a requirement therefore for an air cooled bearing suitable for use in a gas turbine engine having a total loss oil system where the cooling effectiveness of the cooling air is improved in comparison with known arrangements.

According to an aspect of the invention there is provided an air cooled bearing comprising means for imparting a swirl component to the cooling air flowing toward the bearing. By imparting a swirl component to the cooling air the amount of work imparted on the cooling air by the rotating cage and rolling elements of the bearing can be reduced and in some cases substantially eliminated.

In preferred embodiments, the means for imparting a swirl component comprises a vane or a plurality of vanes. The vane or vanes may be fixed in relation to a bearing support or the vanes may be rotatable with respect to the bearing support. Alternatively one or more vanes may be fixed in relation to the bearing support and at least one further vane may be rotatable with respect to the bearing support.

In preferred embodiments the vane or vanes may be connected to the rotatable shaft supported by the bearing, for example an assembly comprising either a "blisk", that is to say an integral bladed disc, or a series of relatively small blades may be attached to the rotatable shaft in the region of the bearing cage.

Preferably, the bearing comprises a cage for retaining rolling elements of the bearing, and whereby the means for imparting a swirl component to the cooling air is configured so that the swirl direction of the cooling air is substantially the same as the direction of rotation of the bearing cage. In this way the vanes pre-swirl (rotate) the cooling air so that when it enters the gap between the cage and the rollers it has a circumferential component in the same direction as the direction of rotation of the cage. Preferably, the cooling air enters the gap between the cage and the rolling elements with a circumferential velocity component substantially the same as the rotational velocity of the bearing cage. In embodiments where the cooling air is pre-swirled in the same direction and at the same speed as the bearing cage the heating effect is minimised,

and the torque acting on the cage due to the work done on the air by the cage is also minimised. In this way the operational temperature of the bearing is reduced and the probability of the bearing rolling elements skidding is also reduced.

In preferred embodiments a plurality of pre-swirl vanes impart a circumferential velocity component on the cooling air that is approximately half the shaft rotation speed, that is to say half the engine speed in a gas turbine application.

The present invention also contemplates a gas turbine engine or lift fan engine or the like comprising at least one air cooled bearing having means for imparting a swirl component to cooling air flowing towards the said bearing.

Preferred embodiments of the invention will now be more particularly described, by way of example only, with reference to the accompanying drawings, in which:

Figure 1 is an axi-symmetric view of an air cooled bearing according to one embodiment and invention;

Figure 2 is a view similar to that of Figure 1 showing a different embodiment of the invention; and

Figure 3 is a view similar to that of Figures 1 and 2 showing a further embodiment of the invention.

Referring to Figure 1, an air cooled bearing assembly 10 for rotatably supporting a gas turbine engine shaft 12 comprises a rolling element type bearing 14 having an inner race 16, an outer race 18 and a plurality of spherical balls 20 positioned radially between the inner and outer races 16 and 18. The balls 20 are retained within a bearing cage structure 22 as is well known in the art. The bearing 14 rotatably mounts the engine shaft 12 with respect to a bearing support structure 24 which is fixed with respect to the engine casing (not shown). The bearing assembly 10 is lubricated by lubricating oil which is fed through one or more oil feed galleries 26 in the bearing

support 24 and the outer race 18. The galleries 26 constitute part of a total loss oil lubrication system.

The bearing assembly 10 is further provided with an annular array of circumferentially spaced vanes 28 which are fixed in relation to the bearing support 24 and extend radially between the bearing support 24 and the engine shaft 12. The vanes 28 constitute pre-swirl vanes for imparting a swirl component in the circumferential direction of the bearing on cooling air flowing towards the bearing, as indicated by cooling flow arrows 30, for the purpose of cooling the bearing assembly 14. The cooling air 30 is preferably bled from the engine compressor and flows from left to right in the drawing of Figure 1 in an axial direction towards the bearing assembly 14. Prior to entering the bearing assembly, in the region between the inner and outer races, the cooling air passes through the array of vanes 28 which act to pre-swirl the air so as to provide the air with a circumferential component about the engine axis.

The amount of swirl, or angle of swirl, imparted on the cooling air by the vanes 28 is determined so that the swirl velocity and direction is substantially the same as the rotational velocity and direction of the bearing cage 22 at engine design speed, that is to say the normal rotational speed of the engine in use. As is well known in the art, the rotational speed of the bearing cage 22 is approximately half that of the rotational speed of the engine shaft 12 supported by the bearing assembly.

In the embodiment of Figure 1 the vanes 28 are fixed relative to the bearing support 24 such that the vanes 28 define an array of stator vanes. In the embodiment of Figure 2 the bearing assembly 10 is identical to that of Figure 1 except that the vanes 28 are fixed in relation to and circumferentially spaced about the engine shaft 12. In this embodiment the vanes 28 comprise an annular array of rotor vanes which rotate about the engine axis with the engine shaft 12. The vanes 28 in the embodiment of Figure 2 function in the same way as the vanes in the embodiment of Figure 1 in the sense that they act as pre-swirl vanes for imparting a swirl component to the cooling air 30 flowing towards the bearing 14. In the embodiment of Figure 2 the vanes 28 extend from the

surface of the shaft 12 radially towards the bearing support structure 24 with the tips of the vanes defining a relatively small clearance gap between the vanes the bearing support 24. In a slightly modified embodiment to that of Figure 2 (not shown) the rotor vanes 28 may be part of an integral bladed disk (blisc or blisk) which is fixed to the engine shaft 12 for rotation therewith.

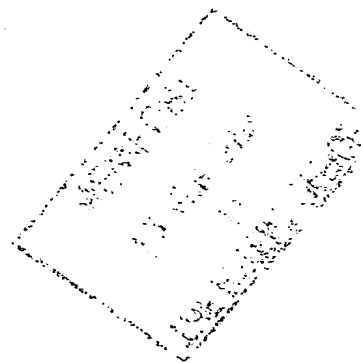
In the embodiment of Figure 3 the bearing assembly 10 comprises an upstream array of circumferentially spaced stator vanes 28a, as in the embodiment of Figure 1, and an array of circumferentially spaced rotor vanes 28b downstream of the stator vanes 28a similar to the arrangement of the stator vanes 28 in the embodiment of Figure 2. The combined effect of the stator and rotor vanes 28a, 28b in the embodiment of Figure 3 is the same as the vanes in the embodiment of either Figures 1 or 2, that is to say to impart a swirl component to the cooling air 30 flowing towards the bearing assembly 14.

Although aspects of the invention have been described with reference to the embodiments shown in the accompanying drawings, it is to be understood that the invention is not limited to those precise embodiments and that various changes and modifications may be effected without further inventive skill and effort. For example, the invention envisages embodiments of an air cooled bearing in applications other than a gas turbine engine, and/or embodiments where the bearing is fed with a limited flow of lubricating oil in a total loss lubrication system. The present invention is applicable to any bearing application where cooling or additional cooling of the bearing is required.

CLAIMS

- 1 An air cooled bearing comprising means for imparting a swirl component to cooling air flowing towards the bearing.
- 2 A bearing as claimed in Claim 1 wherein the said means for imparting a swirl component comprises a vane or a plurality of vanes.
- 3 A bearing as claimed in Claim 2 wherein the said vane or vanes is/are fixed in relation to a bearing support means of the said bearing.
- 4 A bearing as claimed in Claim 2 wherein the said vane or vanes is/are rotatable with respect to a bearing support means of the said bearing.
- 5 A bearing as claimed in Claim 4 wherein the said vane or vanes is/are connected to a rotatable shaft supported by the said bearing.
- 6 A bearing as claimed in Claim 2 wherein at least one vane is fixed with respect to a bearing support means of the said bearing and at least one further vane is rotatable with respect to the said bearing support means.
- 7 A bearing as claimed in any preceding claim wherein the said bearing comprises a cage for retaining rolling elements of the said bearing and wherein the said means for imparting a swirl component to the cooling air is configured such that the swirl direction of the cooling air is substantially the same as the direction of rotation of the said bearing cage in use.
- 8 A bearing as claimed in Claim 7 wherein the said means for imparting a swirl component to the cooling air is configured such that the circumferential swirl velocity of the cooling air is substantially equal to the rotational velocity of the said bearing cage in use.

- 9 A bearing as claimed in any preceding claim wherein the said bearing comprises a total loss oil lubrication means.
- 10 A gas turbine engine comprising at least one bearing as claimed in any preceding claim.
- 11 A bearing substantially as hereinbefore described and/or with reference to the accompanying drawings.
- 12 A gas turbine engine substantially as hereinbefore described and/or with reference to the accompanying drawings.



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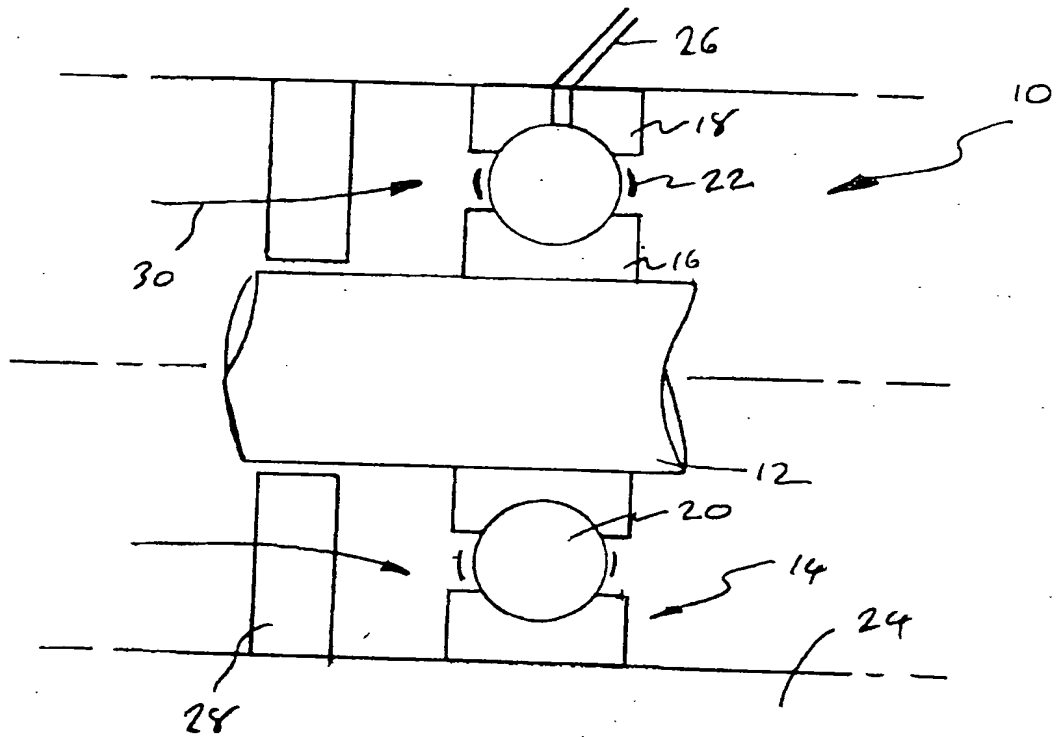


FIG 1

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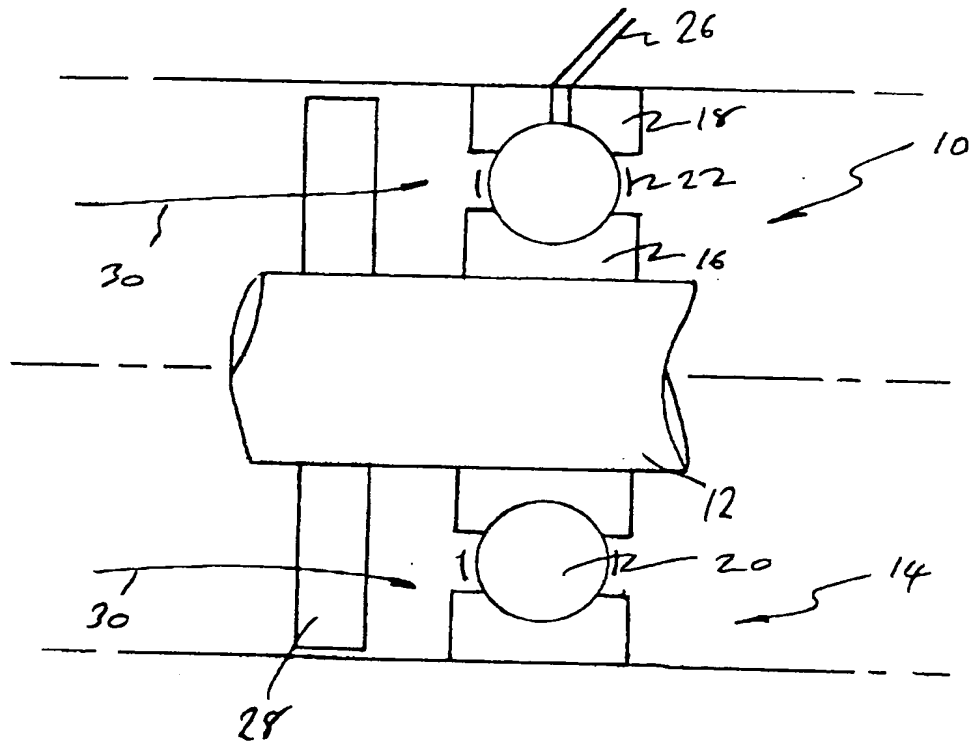


FIG 2

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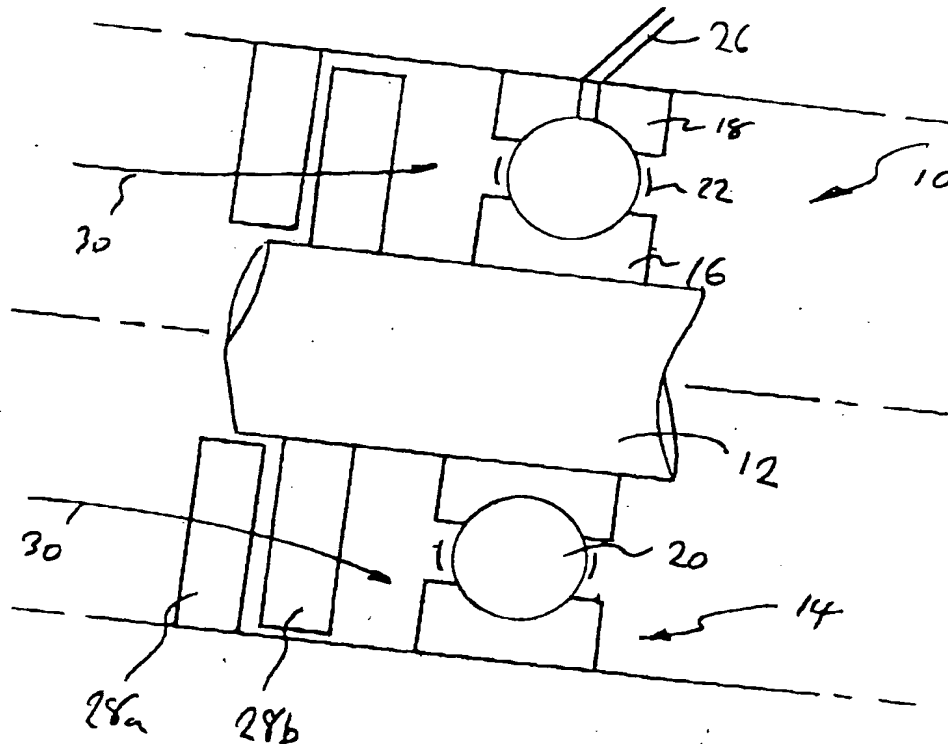


FIG 3

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